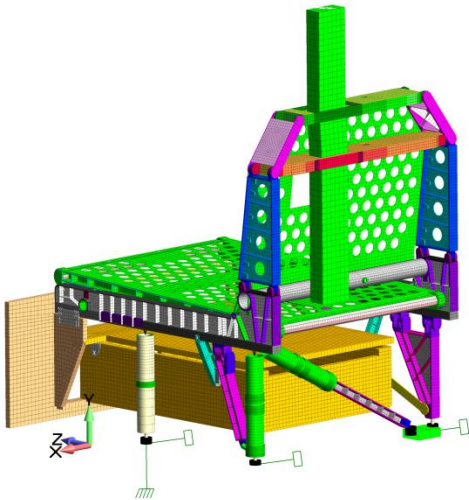


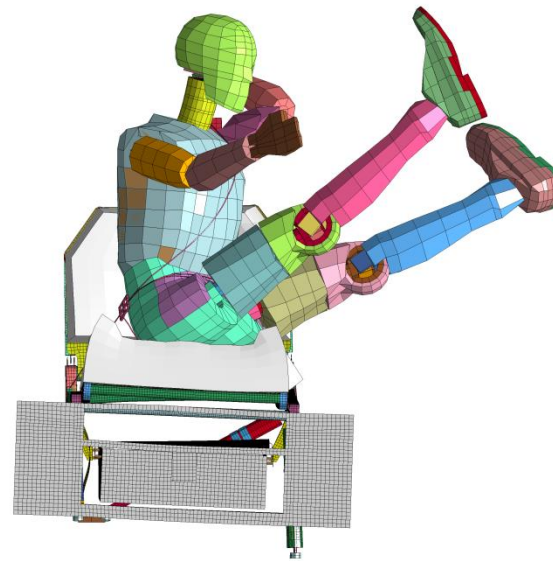


16g Seat Crash Test TSO-C127a / SAE AS8049A / 14 CFR Part 25.562

Femap FEA Model



FEA + LS-DYNA Model



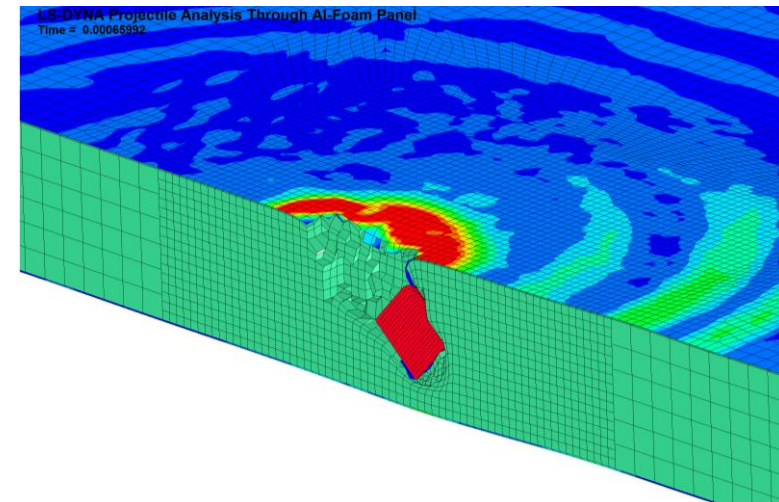
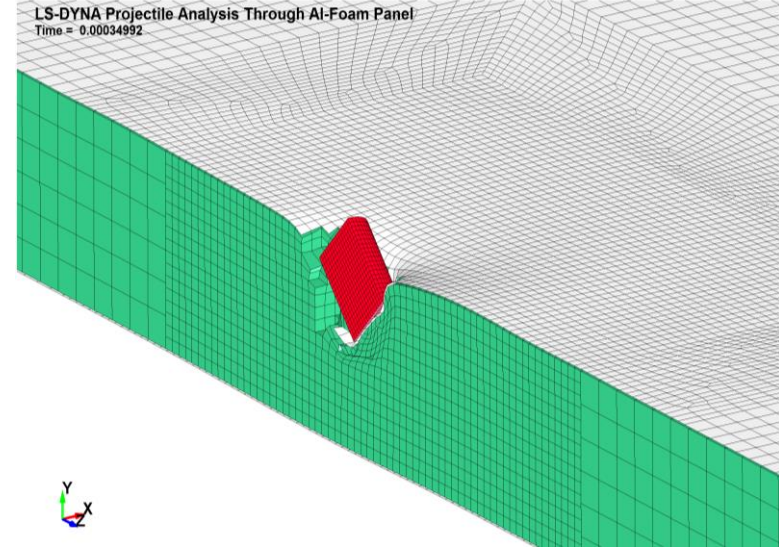
Validation is Gold

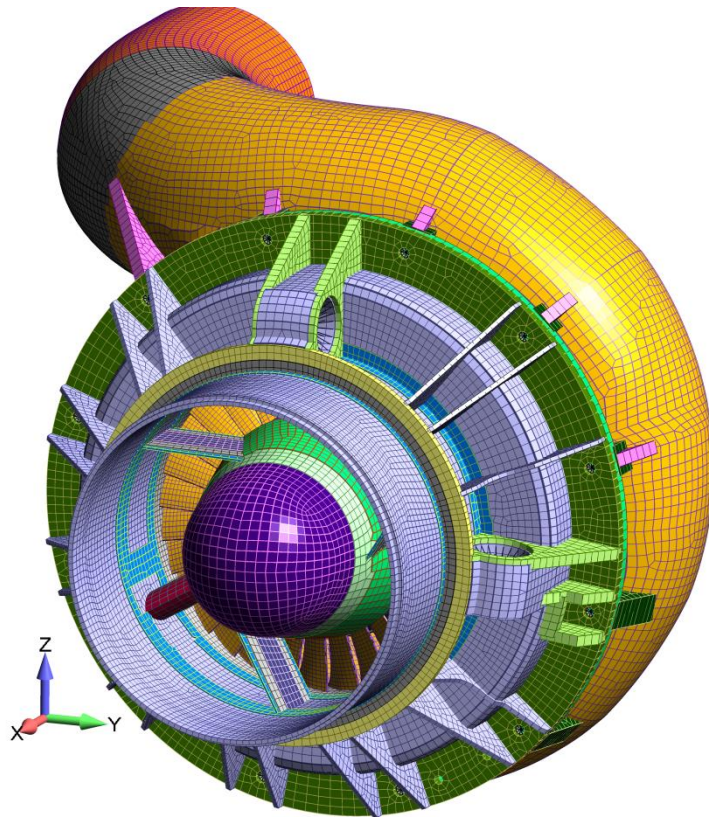


Passing the FAA 16g sled test is no trivial matter for highly optimized aluminum and composite airplane seats. The objective of this LS-DYNA study was to ensure that the client's seat could be validated against the test sled results and that subsequent seat versions would pass "the first time".

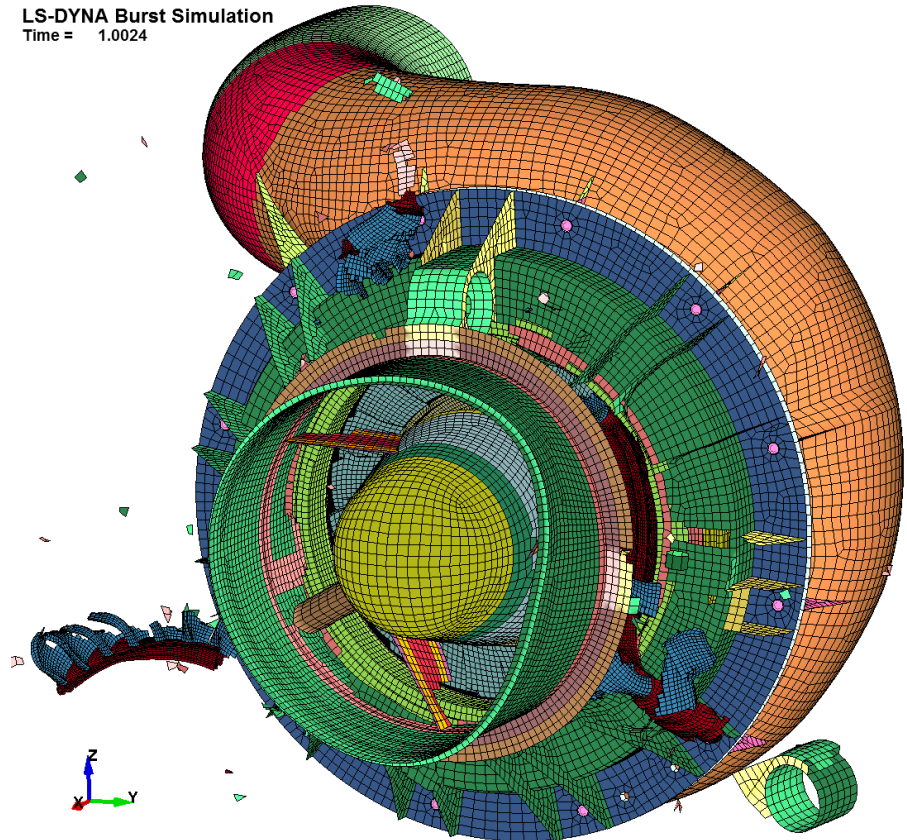


A projectile penetration study was conducted to assess the protective capabilities of a standard aluminum skinned foam sandwich panel. These types of panels are commonly used to create lightweight truck-mounted mobile shelters. For this analysis work, a section of the panel was idealized into a plate and brick FEA model. The panel was subjected to a secondary ballistic impact penetration of a grenade fragment falling at terminal velocity. The final results allowed the client to meet their design requirements without the need of experimental testing.





LS-DYNA Burst Simulation
Time = 1.0024

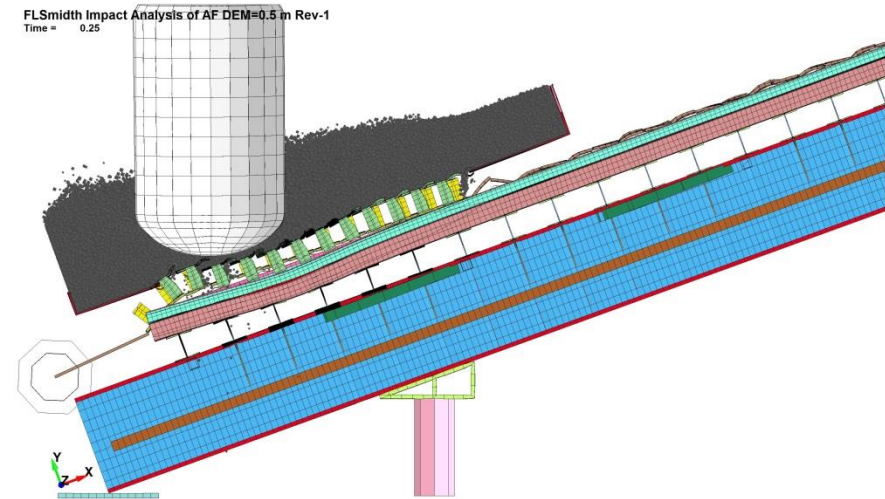
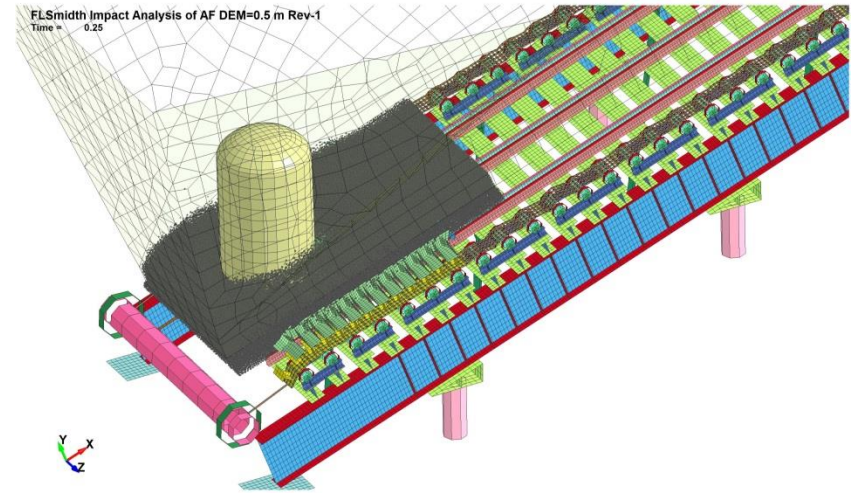


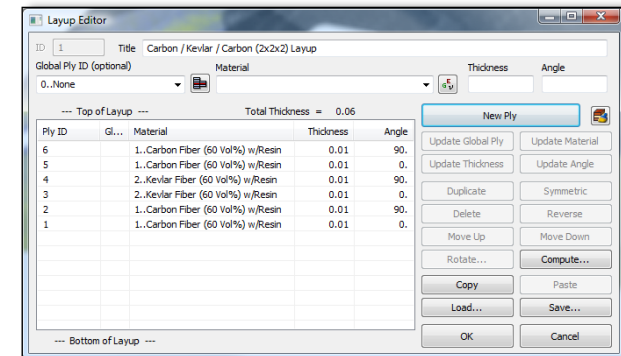
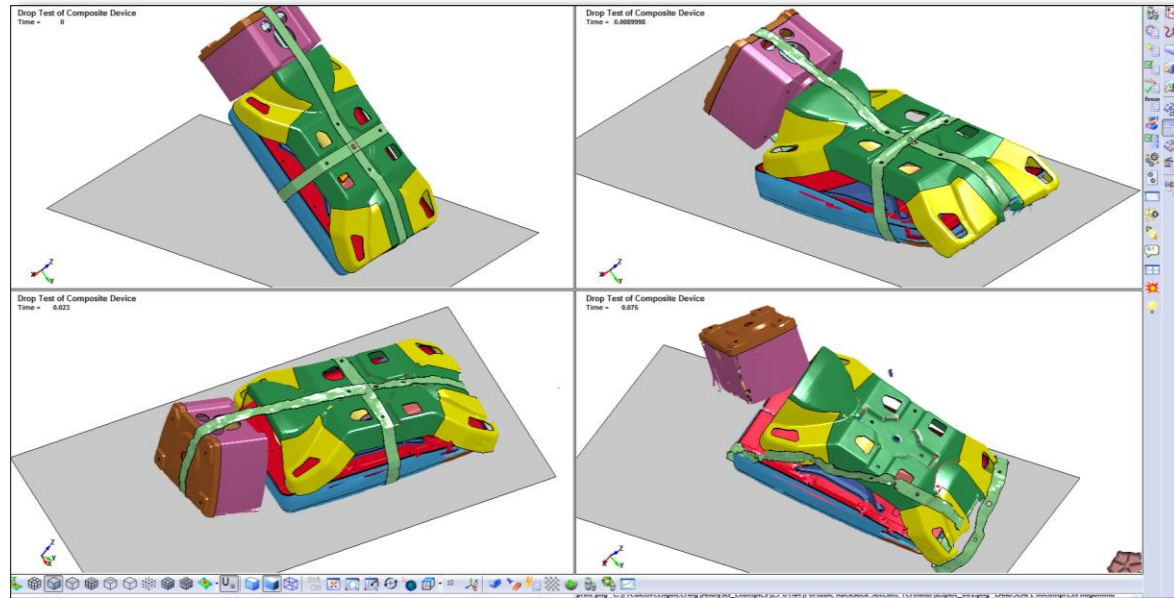
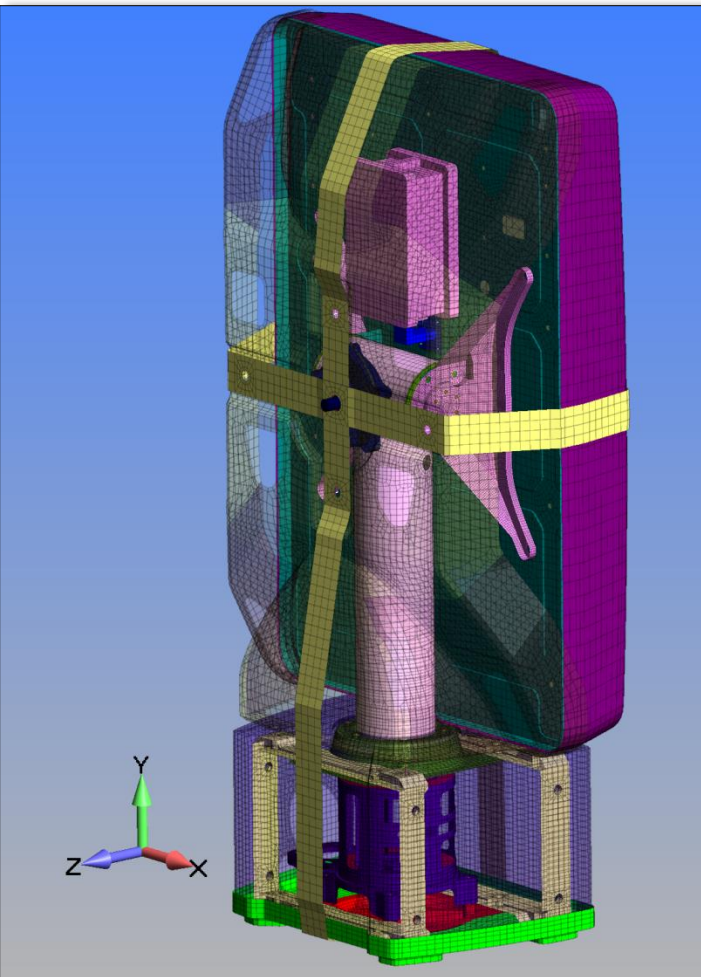
LS-DYNA turbine burst simulation of an air drive power turbine. Analysis work led to significant costs and schedule savings; e.g., each simulation was approximately \$5k and four days as compared to the burst tests at over \$100k and 30 days.

Impact Analysis using Discrete Element Method



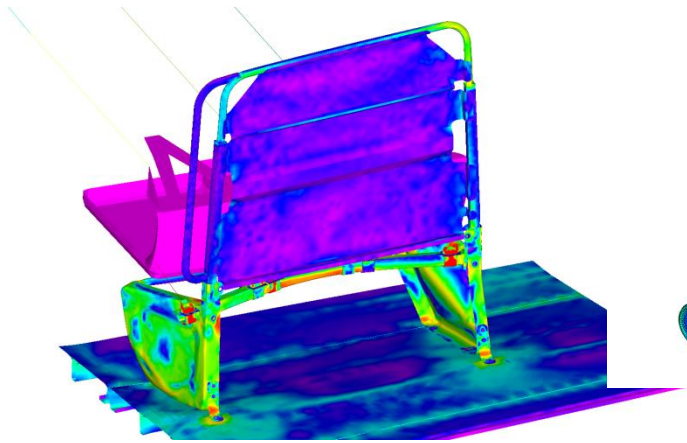
LS-DYNA was used in a combined structural / DEM model for the simulation of a large rock-drop on an apron feeder (AF) commonly used within the mining industry. Results show that if the AF is kept filled with material, the impact of large rocks is almost completely mitigated.



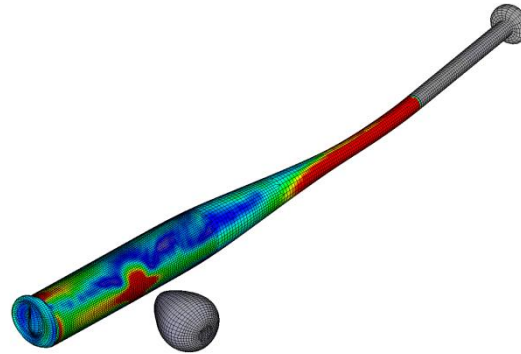


An ultra light-weight carbon fiber composite electronic device was drop tested through a range of 26 positions (MIL-STD 810e). The shell of the unit was a blend of carbon and Kevlar layers for increased impact resistance. The finite element model was used to document experimental drop test failures and then to implement solutions. The modeling results were reviewed by a team of external experts and accepted for production.

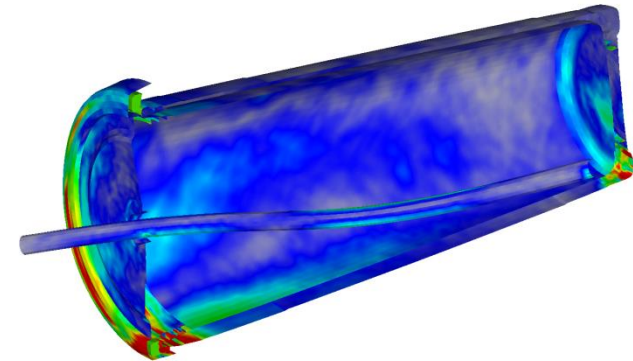
Bus Seat Development



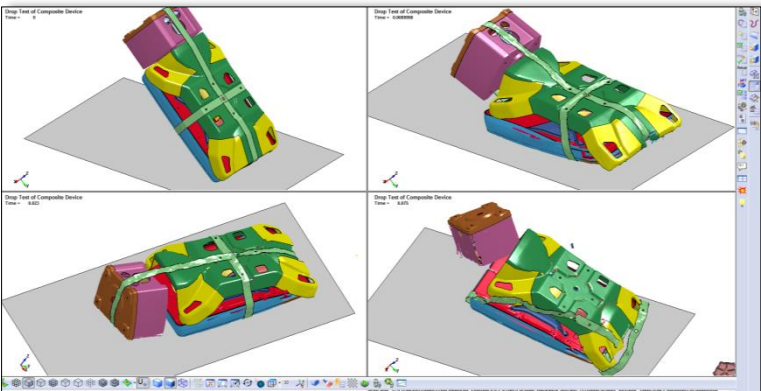
Sports Equipment



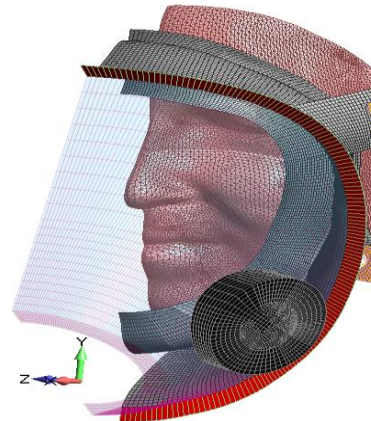
Drop Test of Consumer Products



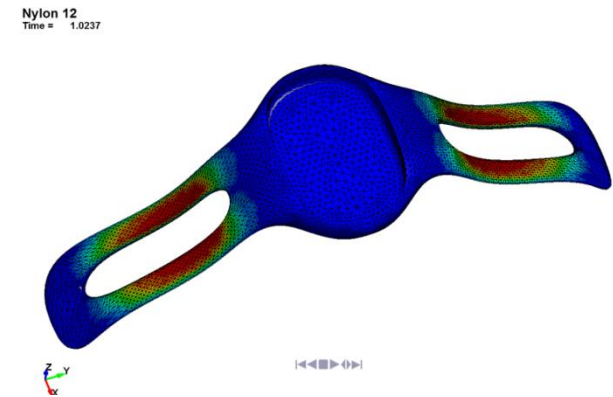
Drop Test of Electronics



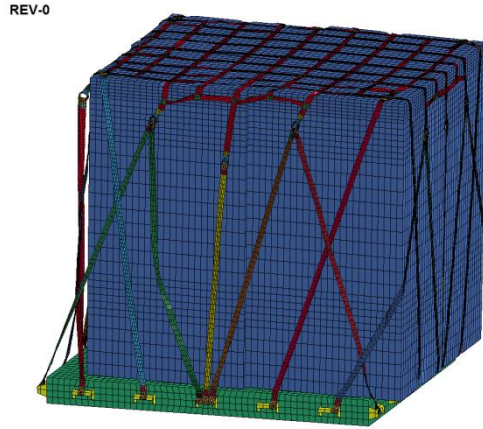
Human Biometrics



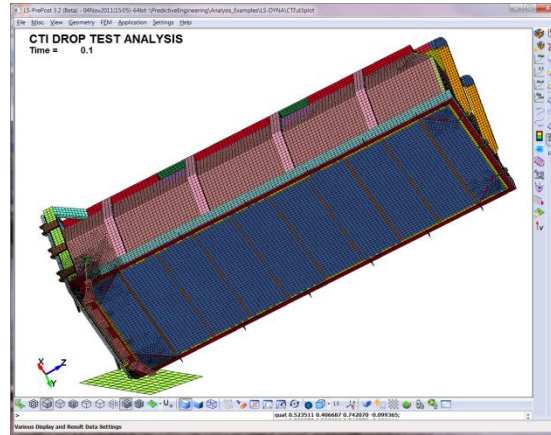
Plastic Structures



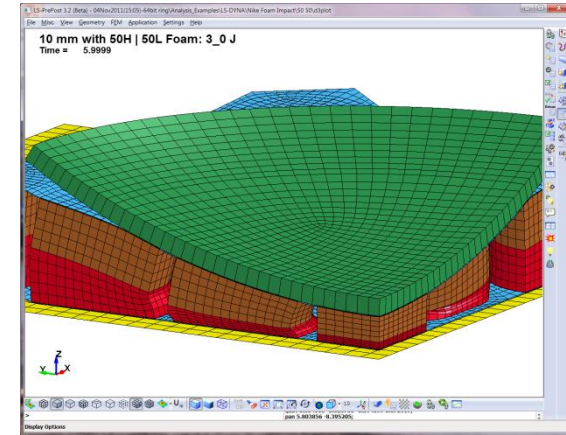
Cargo Net Development



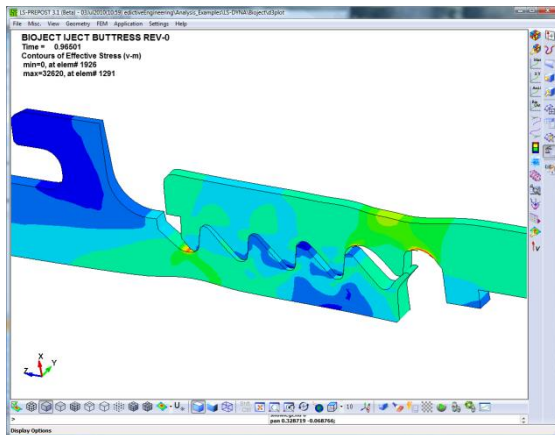
Impact Analysis



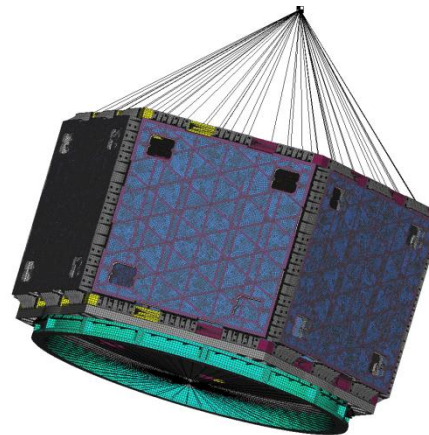
Impact of Plastic Foams



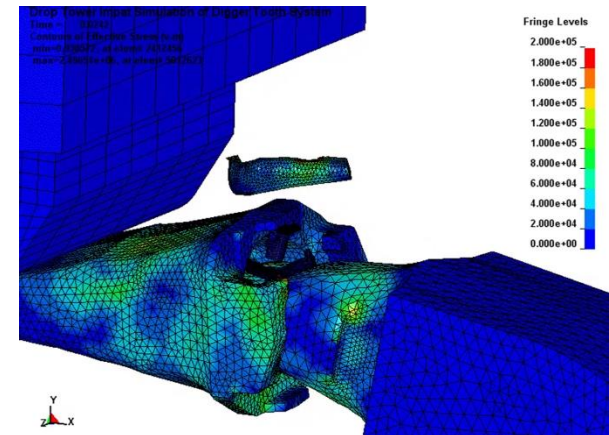
Plastic Thread Design



Modal Analysis



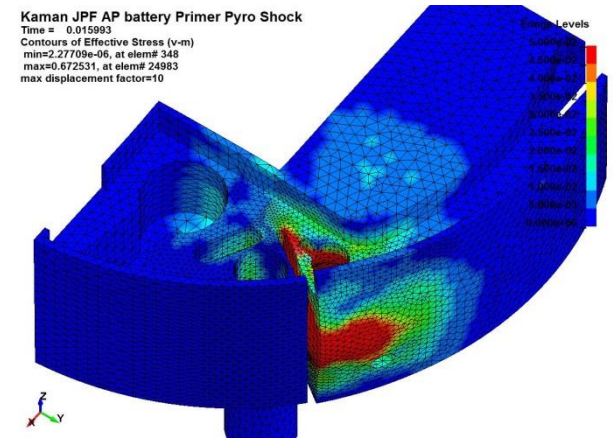
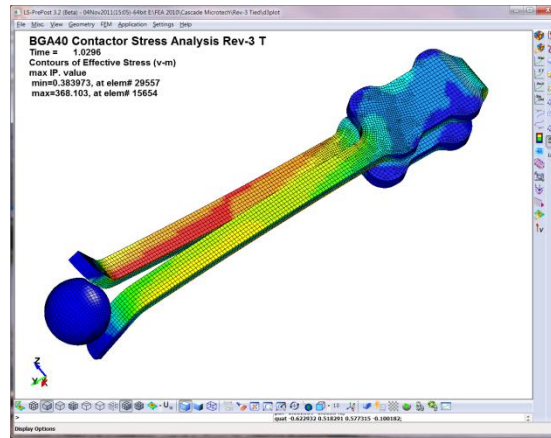
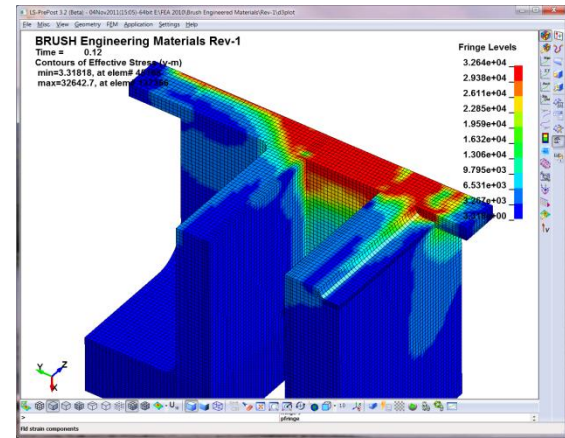
Digger Tooth Failure Simulation



Electron Beam Welding

Elastic-Plastic Contact

Pyro-Shock Analysis

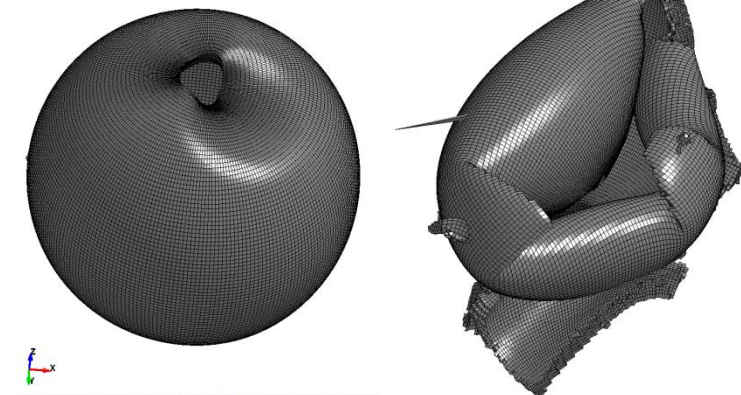


D.O.E.R. Glass Bathysphere

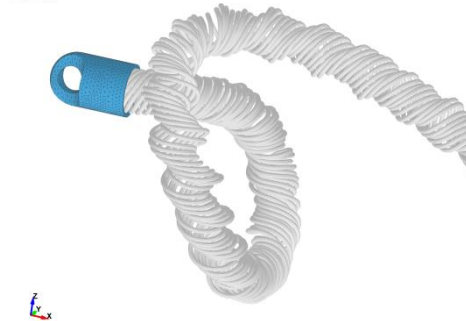
Medical Equipment

Fracture Mechanics

DOER Glass Sphere Buckling Analysis
 Time = 0.18



Bone Marrow Drill
 Time = 0.027

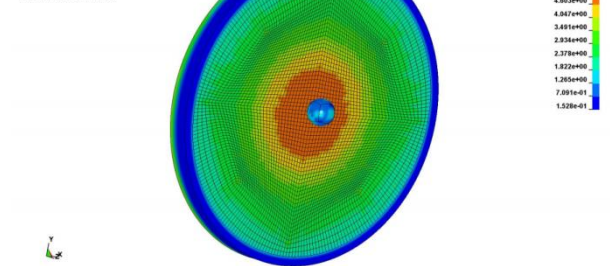


Arnold Engineering Development Corporation
 Training For the Joint Flight

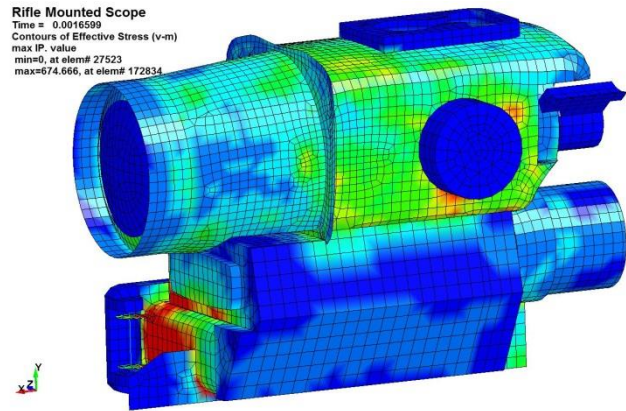
Arnold AFB Wind Tunnel Impact and Fracture Mechanics Analysis Rev-0

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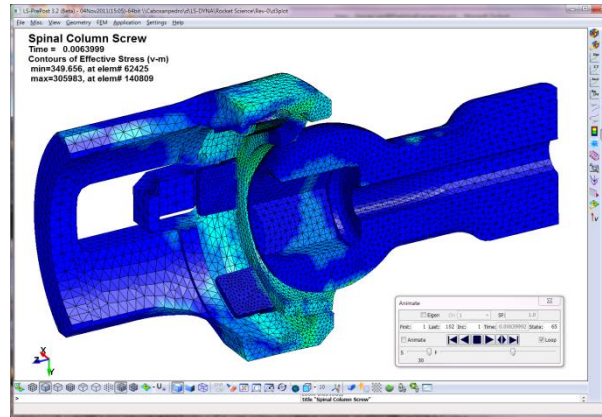
AFB Window Impact Simulation Rev-0_A
 Time = 8.013001



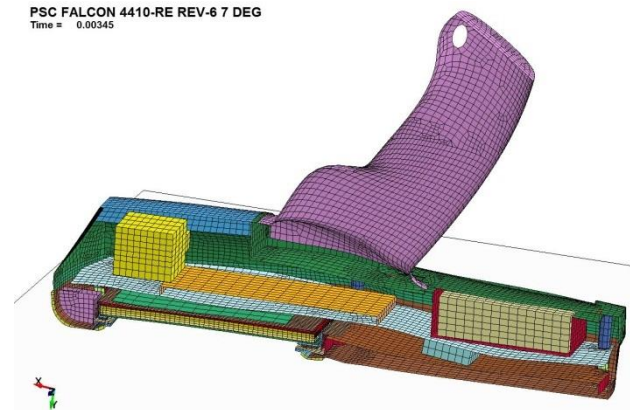
Ballistic Shock Loading



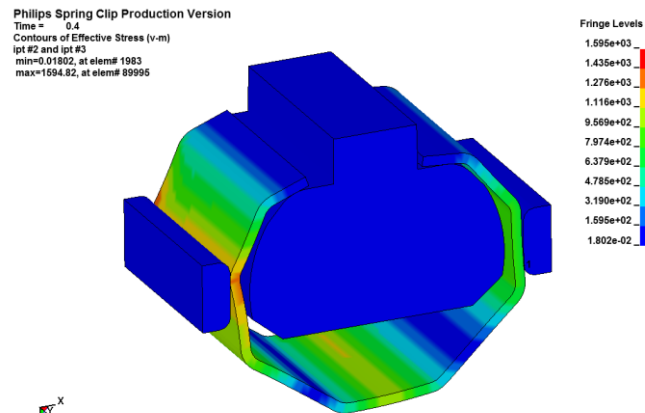
Max. Load Failure Analysis



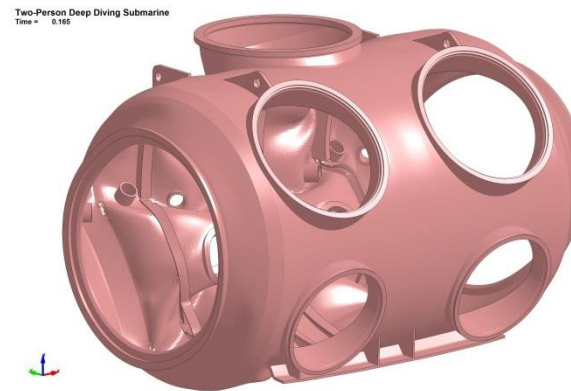
Hand-Held Scanner Drop Test



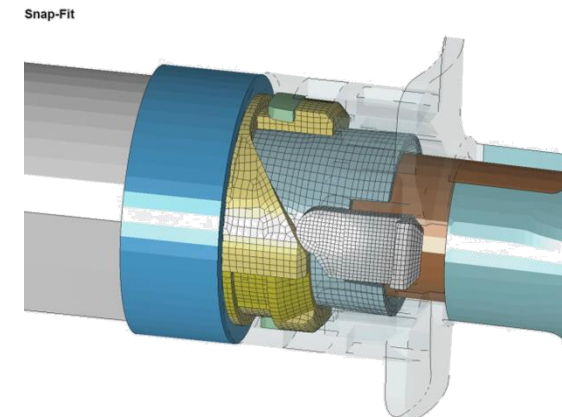
Toothbrush Spring Design



Non-Linear Buckling of Sub

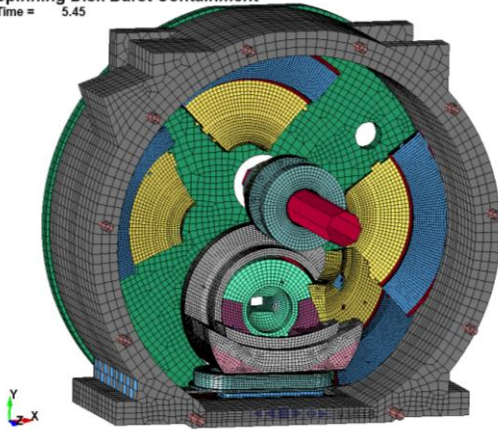


Medical Device – Plastic Snap-Fit



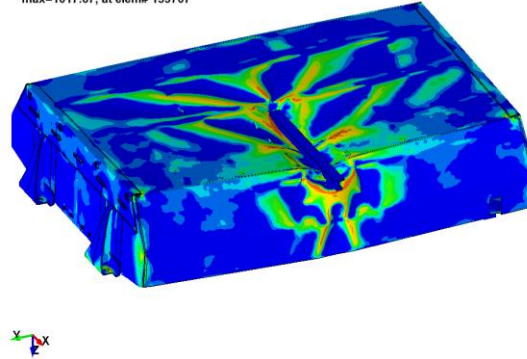
Disk Burst Containment

Spinning Disk Burst Containment
 Time = 5.45



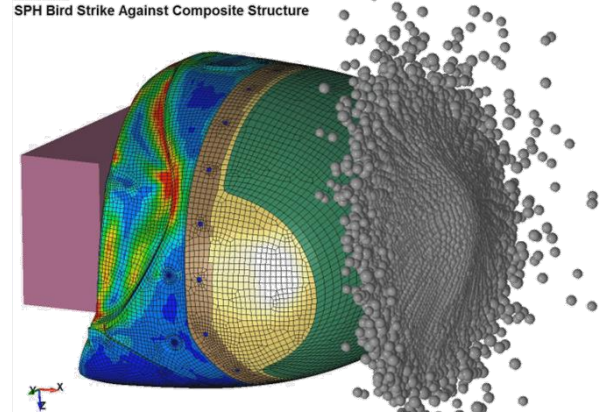
Locomotive Fuel Tank

Locomotive Fuel Tank Crushing Analysis
 Time = 0.050001
 Contours of Maximum Principal Stress
 Ipt #2 and Ipt #3
 min=-0.0048224, at elem# 276069
 max=1017.67, at elem# 139707

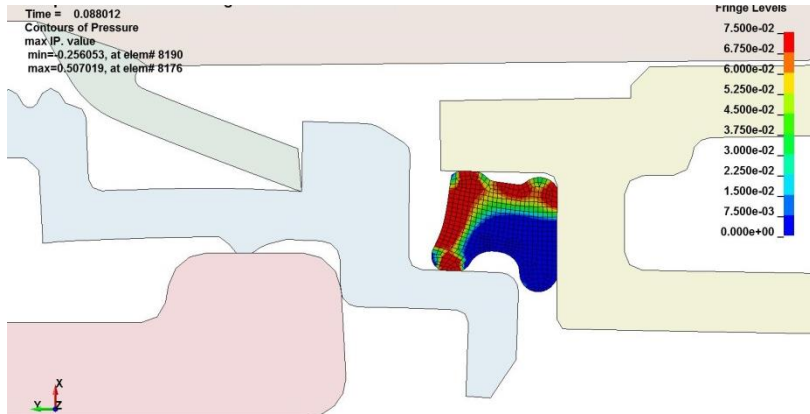


SPH Bird-Strike

SPH Bird Strike Against Composite Structure



Hyperelastic Silicone Seal Design



General LS-DYNA

www.PredictiveEngineering.com
 for additional information on
 projects.